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COMMITTEE ON THE PEACEFUL USES  
OF OUTER SPACE

INFORMATION FURNISHED IN CONFORMITY WITH GENERAL ASSEMBLY  
RESOLUTION 1721 B (XVI) BY STATES LAUNCHING OBJECTS INTO  
ORBIT OR BEYOND

Note verbale dated 1 May 1978 from the Permanent Representative of  
Japan to the United Nations addressed to the Secretary-General

The Permanent Representative of Japan to the United Nations presents his compliments to the Secretary-General of the United Nations and, in conformity with General Assembly resolution 1721 B (XVI), has the honour to transmit herewith information concerning a space object which was launched into geostationary orbit by Japan on 15 December 1977 with the co-operation of the National Aeronautics and Space Administration of the United States of America.

1. Name of satellite: Medium-capacity Communications Satellite for Experimental Purposes (CS)  
(Japanese name: Sakura)
2. International designation: 1977-118A
3. Launching vehicle: Delta Launch Vehicle 2914-137
4. Date and place of launch:
  - (1) Date: 00:47 (u.t.), 15 December 1977
  - (2) Place: Eastern Test Range, Cape Canaveral, Florida, the United States of America
5. Launching organization: National Space Development Agency of Japan (NASDA)  
National Aeronautics and Space Administration of the U.S.A. (NASA)

Note:

NASA furnished spacecraft launching and associated services to NASDA at the request of NASDA on a reimbursable basis.

NASDA injected the CS into the geostationary orbit.

6. Orbital parameters:
  - (a) Period: 1436 minutes (i.e. 23 hours and 56 minutes)
  - (b) Inclination:  $0.11^{\circ}$
  - (c) Perigee: 35,783 Km
  - (d) Apogee: 35,790 Km
  - (e) Geographical longitude on geostationary orbit:  $135^{\circ}$  E

7. General function:

CS is a spin-stabilized geostationary communications satellite. The satellite is equipped with basic systems such as TT and C system (2 GHz band and 6/4 GHz band), electric power supply system and so on, and mission equipment such as two transponders of 6/4 GHz band, six of 30/20 GHz band and so on. Following experiments are to be carried out.

- (1) Measurement of on-board mission equipment characteristics
- (2) Measurement and evaluation of propagation characteristics, especially, in quasi-millimetric waves
- (3) Experiments on signal transmission through satellite communication system

(4) Experiments on satellite communication system operation

(5) Experiments on satellite operation and control

8. Characteristics of satellite:

(1) Weight: approximately 340 kg (at an early stage in orbit)

(2) Physical configuration and dimensions:

(a) Configuration: Cylindrical satellite

(b) Height: 3.48 m

(c) Diameter: 2.18 m

(3) Attitude control subsystem: Spin stabilization

(4) Expected life: More than three years

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