

### JAXA's Contribution to Space Weather; Arase(ERG) and other satellites

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## INTRODUCTION

- ✓ JAXA (The Research and Development Directorate : RRD) has measured the space radiation environment since 1987.
- Our purpose: Investigation possible relationship between satellite failure and space radiation environment
- ✓ The Kiku-5 (ETS-5) was our first satellite in geostationary Earth orbit (GEO) to measure electrons, heavy ions and protons.

Launch Date	Aug 27,1987 (~ Sept 27,1997)
Location	Tanegashima Space Center
Orbiter	GEO
Altitude	About 36,000km (east longitude 135°)
Inclination	0 degree
Period	24 hours





## TEDA AND SEDA

- TEDA and SEDA are instruments that have been developed by JAXA to measure space environment. (high energy particles, galactic cosmic rays, atomic oxygen, plasma, magnetic fields, etc.)
- $\checkmark$  We have 16 types of instruments.
- TEDA and SEDA have been installed on seventeen spacecraft. Eight spacecraft are now in operation.



#### Space Environments Satellites

# TEDA AND SEDA

$\frown$	Equipment	Specifications
V	DOse Monitor (DOM)/Standard DOse Monitor (SDOM)	Electrons: 0.5 - 50 MeV
		Protons: 0.9 - 250 MeV
$\frown$		Heavy ions: 6.7 - 270 MeV
2)	High energy Particle Monitor (HPM)	Protons:
$\overline{\mathbf{A}}$		50 - 70 MeV/70 - 100 MeV/100 - 180 Mev
3	Heavy Ion Telescope (HIT)	Li, Be, B: 10 - 23 MeV
		C, N, O: 15 - 36 MeV
$\frown$		Fe: 31 - 76 MeV
4/	DOSimeter (DOS)	V <sub>gs</sub> at Id = 1 mA, 10 mA, 100 uA and 10 uA
5	MAgnetoMeter (MAM)	Range-H:±256 nT
$\sim$		Range-L:±65536 nT
6	Potential Monitor (POM)	Surface material
		OSR/Kapton evaporated Al/Teflon evaporated Ag/Metallic
		board insulator. Voltage: +5 kV to -10 kV
7	Atomic Oxygen Monitor (AOM)	10 <sup>10</sup> to 10 <sup>13</sup> atoms/m <sup>3</sup>
8	NEutron Monitor (NEM)	Bonner Ball type Neutron Detector (BBND)
		Neutrons: 0.025 eV - 15 MeV
		Scintillation Fiber Detector (FIB)
9	PLAsma Monitor (PLAM)	Density and electron temperature of space plasma
10	Dlscharge Monitor (DIM)	Discharge •Specification
		0.75 to 58 V
(11)	Single event Upset Monitor (SUM)/RAM Soft-error Monitor	Monitoring errors by single event.
	(RSM)	
12	Integrated Circuit Monitor (ICM)	Measuring degeneration and change of IC.
13	Solar Cell Monitor (SCM)	Researching the properties of solar cells in the space
		environment.
(14)	COntamination Monitor (COM)/Thermal control material	Measuring the solar absorption and infrared emission of
	Degradation Monitor (TDM)	samples.
15	Light Particle Telescope (LPT)	ELS-A: Electrons: 0.03 MeV - 1.3 MeV
		ELS-B: Electrons: 0.3 MeV - 20 MeV
		APS-A: Protons: 0.4 MeV - 33 MeV, α1.5 MeV - 250 MeV
		APS-B: Protons: 0.7 MeV - 40 MeV, α2.5 MeV - 100 MeV
16	Extremely High-Energy Electron Experiment (XEP)	Electrons: 0.4 MeV - 20MeV

	Satellite name Operational period		Orbit	Equipment
I. Geostationary Earth Orbit (GEO Altitude: 36,000 km)				
1	ETS-V	1987/8/27 - 1997/9/12	Altitude: approx. 36,000 km Inclination:0 degrees	DOM/POM/DIM RSM/ICM/SCM/TDM
2	2 DRTS 2002/9/10 -		Altitude: approx. 36,000 km Inclination: 0 degrees	SDOM
3 ETS-VIII		2006/12/18 - 2017/1/10	Altitude: approx. 36,000 km Inclination: 0 degrees	DOS/MAM/POM/ SUM
4	Michibiki(QZS) 1 <sup>st</sup> and 2 <sup>nd</sup>	2010/09/11 -	Altitude: Perigee 32,600 km, Apogee 38,950 km Inclination: approx. 41 degrees	LPT/POM(only 1 <sup>st</sup> )/MAM
		II. Geosynchronous Transfer	Orbit (GTO Altitude: 250 to 40,000 km)	
5	ETS-VI	1994/8/28 - 1996/1	Altitude: Perigee 8,600 km Apogee 38,600 km Inclination: approx. 13 degrees	DOM/HIT/MAM POM/SUM//ICM/ SCM/COM
6	MDS-1	2002/2/4 - 2003/9/25	Altitude: Perigee 209 km Apogee 35,204 km Inclination: 29.1 degrees	SDOM/HIT/DOS/ MAM/SUM
7	ARASE(ERG)	2016/12/20 -	Altitude: Perigee: approx. 300 km Apogee: approx. 33,200 km Inclination: approx. 31 degrees	XEP
	-	III. Sun-synchronous Sub-rec	urrent Orbit (LEO Altitude: 600 to 900 km)	
8	ADEOS	1996/8/17 - 1997/6/30	Altitude: approx. 800 km Inclination: 98.6 degrees	HPM/HIT/DOS POM/SUM/COM
9	ADEOS-II	2002/12/14 - 2003/10/31	Altitude: approx. 803 km Inclination: approx. 99 degrees	DOM/DOS/SUM
10	ALOS	2006/1/24 - 2011/5/12	Altitude: approx. 690 km Inclination: approx. 98 degrees	LPT/HIT
11	GOSAT	2009/1/23 -	Altitude: approx. 667 km Inclination: approx. 98 degrees	LPT/HIT
		IV. Su	ub-recurrent Orbit	
12	JASON-2 (CNES)	2008/6/20 -	Altitude: approx. 1,336 km Inclination: approx. 66 degrees	LPT
13	JASON-3 (CNES)	2016/1/17 -	Altitude: approx. 1,336 km Inclination: approx. 66 degrees	LPT
V. Low Earth Orbit (LEO Altitude: < 600 km )				
14	ETS-VII	1997/11/28 - 1999/12/16	Altitude: approx. 550 km Inclination: approx. 35 degrees	AOM
15	Shuttle/Mir	1998/1/23 - 1998/1/26	Altitude: approx. 400 km Inclination: approx. 51.6 degrees	BBND
17	ISS	1998/1/23 - 1998/1/26	Altitude: approx. 400 km Inclination: approx. 51.6 degrees	BBND
16	ISS (SEDA-AP)	2009/8/24 -	Altitude: approx. 400 km Inclination: approx. 51.6 degrees	SDOM/HIT/AOM/ NEM/PLAM/SUM

#### SPACE ENVIRONMENT AND EFFECTS SYSTEM (SEES)

✓ SEES is a database system for our space environment data.
 ✓ We provided real time and quasi-real-time data through SEES web site (<u>http://sees.tksc.jaxa.jp</u>) for external users

#### <u>How to find SEES web site??</u> [SEES JAXA] and search button click!







Please fill in the IP Address of the computers which a user uses mainly.(Max. 4 computers) Only the access from an IP Address that had you input is admitted. It is not available from the network

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New!! MD

Orbiter: Geo transfer orbi Altitude: Pe Apogee 35,2 Inclination: Operation 2002/2/4 - 20

SDOM: Stando MAM: MAgne

Instruments Spacecraft Data

#### Report

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Notice

## RECENT EVENTS

- Recently the Arase satellite (ISAS/JAXA) and the Japanese GPS satellite Michibiki (QZS) 2nd (Japanese Cabinet office) were successfully launched, and instruments are installed in these satellites.
- ✓ In future the QZS 4<sup>th</sup> (Japanese Cabinet office) will be launched.





- > ARASE(ERG) was launched on December 20, 2016
- Purpose: It is intended for use in elucidating how highly charged electrons are created as they appear and vanish repeatedly along with space storms caused by the disturbance of solar wind, and how space storms develop. It has eight instruments.
- XEP, HEP and MGF has Space Weather data.

Orbital Altitude	Perigee: about 300 km, Apogee: about 33,200 km		
Orbital Inclinarion	about 31°		
Type of Orbit	Elliptical orbit		
Orbit Period	about 538 min.		



Quasi-Zenith Satellite System (QZSS)



- > QZS 2<sup>nd</sup> (Cabinet Office)was launched on June 1, 2017
- Purpose: Japanese GPS second satellite
- QZS 1<sup>st</sup> was launched on September 11, 2010.
- QZS 1<sup>st</sup>, 2<sup>nd</sup> and 4<sup>th</sup> have LPT (radiation particles) and MAM.
- QZS 4<sup>th</sup> will be launched soon.
- QZS 1<sup>st</sup> and 2<sup>nd</sup> space weather data graph will be published soon through SEES.

Orbital Altitude	32000~40000km		
Orbital Inclinarion	about 40°		
Type of Orbit	Quasi-Zenith Orbit		
Orbit Period	about 23 h56 m.		



through

## **ARASE(ERG) SATELLITE**

- ✓ The Arase has nine instruments.
- ✓ 2 instruments for ions (MEP-i/LEP-i) and 4 instruments for electrons (XEP/HEP/MEP-e/LEP-e).
- $\checkmark$  1 instrument for magnetic field (MGF) and 1 instrument for electric field (PWE).
- ✓ XEP, HEP and MGF provide space environment data (quasi-real-time: update every minute).



### MGF/HEP/XEP



#### SPACE WEATHER DATA(MGF/HEP/XEP)

We started to provide space weather data form the SEES/JAXA web site on March 20, 2017



- An organization which send application form can use numeric adia metrix metric (see > application form).
- We provide txt files in quasi-real-time(update every minute).
- ✓ Purpose : We expect them to be used for study of space weather (Prediction).
- ✓ This data is uncalibrated, so users can not use them for science. these data to be used for space weather study.
- ✓ If you are interested in them, please ask us. (SEES > your message) or send us application form (SEES > application form).

Many domestic and international organization/University started using our data!

### **OBSERVATION (ARASE SATELLITE)**

20 2

- ch1

- ch2

- ch3

- ch1

- ch3

ch0 Log fl 6.0 3

3.5

2.2 / 00 1.0 9 ch3 Log flu

7.0 Aay/8/

4.0 cmo/ 601

06 Apr. 2017

12 18

(C) JAXA

05 Apr. 2017

06 12

Time(3 days)

- The March 27 storm: It is caused by the arrival of the high-speed  $\checkmark$ coronal hole stream, accompanies a large increase of the relativistic electrons.
- ✓ The April 4 storm: It has a rapid Dst development and recovery, shows less acceleration and does not recover to the pre-storm level.



### YOUR REQUEST AND QUESTION (ALL JAXA'S SPACE ENVIRONMENT DATA)

- $\checkmark$  How to contact with us
- (1) SEES > your message
- 2 Please send us your mail.(<u>sees@jaxa.jp</u>)
- ✓ How to get an application form (ARASE space weather data).
   ① SEES > application form

SEES	Space Envi	ronment & Effects System	The user registration is here.	JAXA	<b>SEES</b>	Space Environment & Effects System
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	Phone FAX E-mail* Message* Submit Rese		^ 		Application Form	<ul> <li>Address         Research Unit I(SEES system), Research and Development Directrate         Japan Aerospace Exploration Agency (JAXA)     </li> <li>E-mail         sees@jaxa.jp         Tsukuba Space Center.         2-1-1, Sengen, Tsukuba-shi,         Ibaraki-ken, 305-8505 Japan     </li> <li>Fax. (81)29-868-5969</li> </ul>

#### Page to send message

#### Page to application form



## CONCLUSION

- ✓ JAXA(RRD) has measured the space radiation environment since 1987 and it has a database system called 'SEES'
- Recently the Arase satellite and the Japanese GPS satellites Michibiki 2<sup>nd</sup> were launched, so we are getting many space environment data in real-time.
- We have provided mainly real time graph for external users without any limitation, and it is very useful to know space environment.
- ✓ Arase satellite data is the first attempt to provide numeric data file in quasi-real time to an organization/ University.
- But...Arase space weather data is uncalibrated, so users can not use them for science.
- We expect to use these our data for study of space weather nowcast as well as forecast.



### THANK YOU FOR YOUR ATTENTION !



#### [Reference] Particle accelerator @ Tsukuba Space Center in Japan

We have own particle accelerator to calibrate our instruments. This can irradiate to a instrument by low flux electrons/protons. If you are also interested in this, please tell us. We might help calibration of your instrument(radiation particles).



Accelerating voltage	0.4 MeV~2.0MeV
Particle	Electron, Proton
Beam current	lfA~10nA
Beam area	60mmX60mm (max)



# ※ TEDA:TEchnical Data Acquisition equipment ※ SEDA: Space Environment Data Acquisition equipment