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Committee on the Peaceful Uses of Outer Space

Information furnished in conformity with the Convention on Registration of Objects Launched into Outer Space

Note verbale dated 21 August 2003 from the Permanent Mission of the Czech Republic to the United Nations (Vienna) addressed to the Secretary-General

The Permanent Mission of the Czech Republic to the United Nations (Vienna) presents its compliments to the Secretary-General of the United Nations and, in accordance with article IV of the Convention on Registration of Objects Launched into Outer Space,* has the honour to transmit registration data for the Czech satellite MIMOSA, which was launched on 30 June 2003 (see annex).

* General Assembly resolution 3235 (XXIX), annex.

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Annex

Basic registration data for the Czech satellite MIMOSA*

States that launched the object:	The Czech Republic, together with the Russian Federation. The launch was carried out by Eurockot Launch Services GmbH, as part of the Multiple Orbit Mission (MOM), using the Russian Rokot launcher. The Micro-Measurements of Satellite Acceleration (MIMOSA) satellite was manufactured by Space Devices Ltd. of Prague under a contract with the Astronomical Institute of the Academy of Sciences of the Czech Republic. The satellite control and telemetry receiving station at Panska Ves in northern Bohemia belongs to the Institute of Atmospheric Physics of the Academy of Sciences.
Designation of the space object and its registration number:	MIMOSA, 2003-031B (Committee on Space Research designation)
Date of launch and territory from which it was launched:	30 June 2003 at 1415 hours UTC, from the territory of the Russian Federation (Plesetsk Cosmodrome)
Initial orbital parameters:	
Nodal period:	96.31 minutes
Inclination:	96.82 degrees
Apogee:	848 kilometres
Perigee:	319 kilometres
General function of the space object:	The MIMOSA spacecraft is designed for the very precise measurement of atmospheric drag and other forces of non-gravitational origin (such as solar radiation pressure and pressure of terrestrial infrared radiation), which are perturbing the motion of artificial satellites. The satellite, with a total mass of 51.3 kg, has a single scientific instrument, a microaccelerometer, which can monitor changes in satellite acceleration with high precision (10^{-10} metres/second^2). For this type of measurement, it was necessary to design a unique microsatellite

* The registration data are reproduced in the form in which they were received.

with a very regular shape (dimensions: 560x560x570 mm) and install the microaccelerometer at its centre of gravity with extremely high precision. No orbital manoeuvring is envisaged during the mission. There are low-power technical means for satellite telemetry and telecommand using omnidirectional antennae. The expected operational lifetime is 18 months. Received data will be used to develop a detailed model of the density of the upper atmosphere and its changes.